



FIG. 11. Polar curves (lift coefficient  $C_L$  vs. drag coefficient  $C_D$ ) of the Dornier Do. 27 aircraft ( $C_L$  and  $C_D$  values derived from full scale flight tests).  
(By courtesy of the Dornier-Flugzeugwerke.)

a: *Glide* (engine fully throttled, no airscrew thrust) slats open and flaps set at  $0^\circ, 15^\circ, 30^\circ, 45^\circ$ .

b: *Power-on*, throttle fully open, slats open and flaps set at  $0^\circ$  (neutral),  $15^\circ, 30^\circ, 45^\circ$ .